

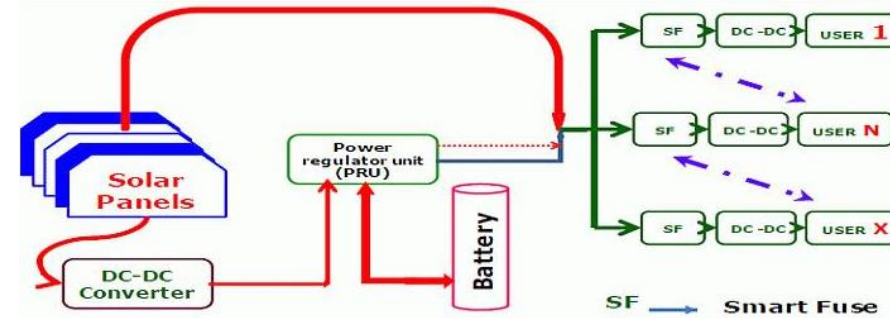


# SATELLITE SUBSYSTEMS

A complete satellite consists of four subsystems:

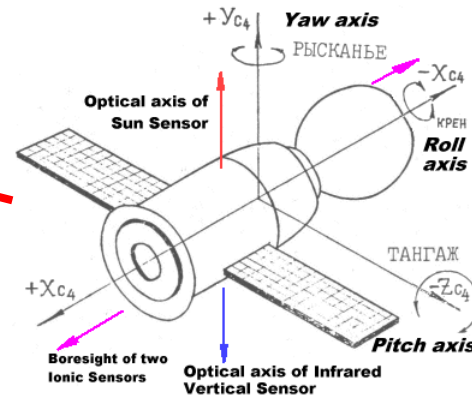
## 1. The power Subsystem consisting of:

- (a) Solar Array
- (b) Battery
- (c) Power supply control unit



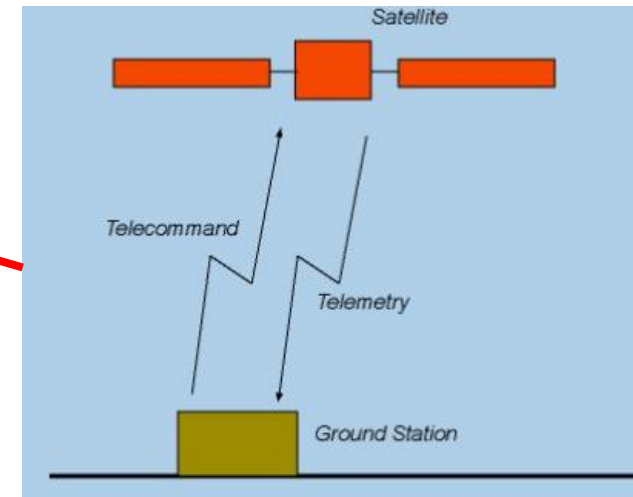
## 2. Attitude and Orbit Subsystem consisting of:

- (a) Sensors
- (b) Propulsion System
- (c) Attitude Control
- (d) Orbit control



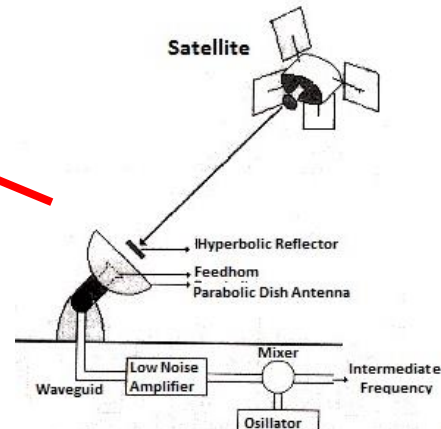
## 3. Telemetry, Tracking and Command Subsystem

- (a) Telemetry system
- (b) Tracking and ranging subsystem
- (c) Command Subsystem



## 4. Communication Subsystem

- (a) Transponders
- (b) Antennas



# 1. POWER SUBSYSTEM

1. Satellite systems use **Solar Cells** or **Radioisotope thermoelectric generators** to supply energy.

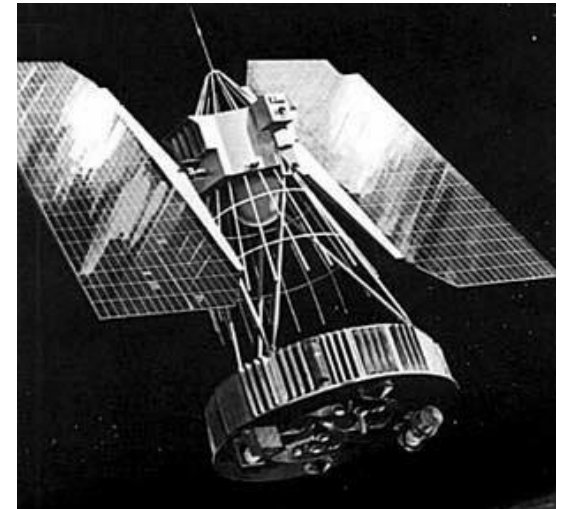
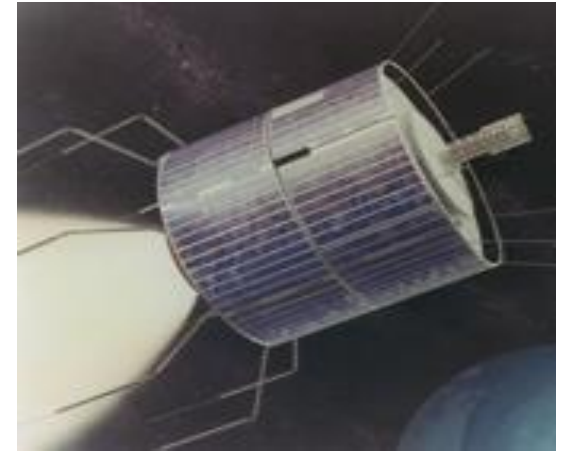
## (a) Solar Cells:

- i. Solar Cells are photovoltaic cells that convert solar radiation to light with **an efficiency of 12-15%**.
- ii. On a typical station, **the solar cells will generate 1-2KW of power**.
- iii. The **number of solar cells on typical satellite may be as many as 1000** covering most of the surface of the satellite.
- iv. If the surface area is inadequate, then the **satellite is equipped with solar panels**.
- v. The efficiency of the solar panels usually **decreases by approximately 30-35% in seven years**.
- vi. Rechargeable batteries are installed and are useful during the satellite eclipse period.

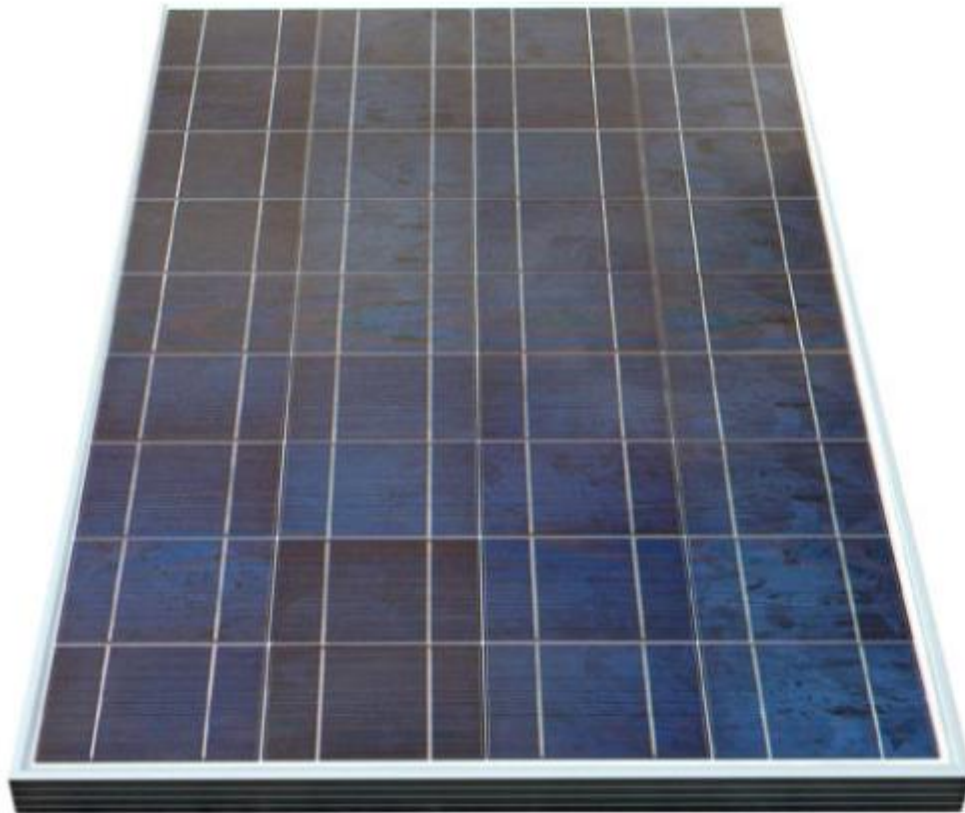
## (b) Radioisotope Thermoelectric Generators:

- I. **They convert heat from decaying radioisotope** (usually plutonium) directly into electrical power
- II. They are **only about 7% efficient**
- III. **93% of the heat is lost or can be used to heat a cold part of the satellite**

2. The **Communication subsystem consumes most of the power**.



# PRICES OF CONSUMER SOLAR PANELS IN KENYA



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\*price depends on the number of panels

**Availability:** 8 - 12 weeks

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**Configuration**

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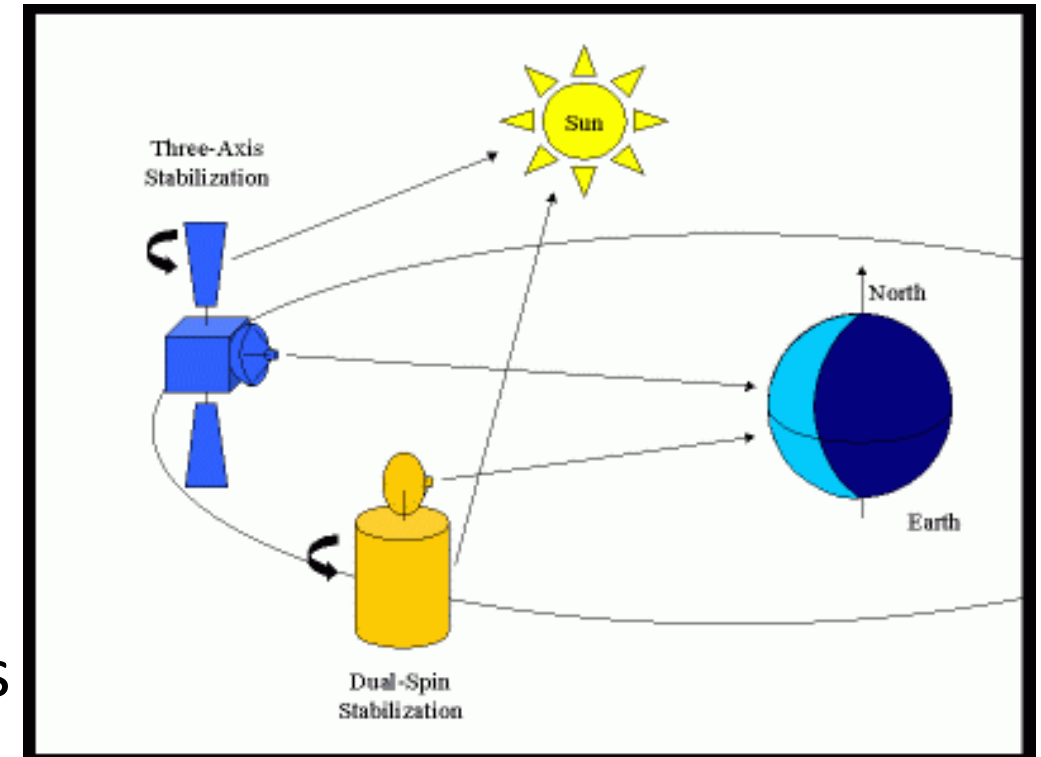


<https://www.cubesatshop.com/product/single-cubesat-solar-panels/>

## 2. SATELLITE STABILIZATION

There are two methods to stabilize a satellite in space, i.e

1. **Dual Speed Stabilization** where the satellite takes the shape of a cylinder which rotates about its long axis.
2. **Three-axis stabilization** where the spacecraft attitude is maintained through the use of momentum wheels or control moment gyros.

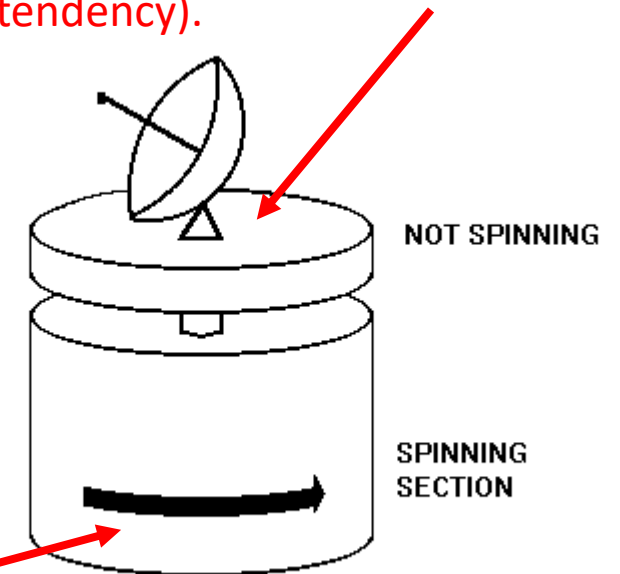


# 2.1 DUAL SPIN SATELLITE STABILIZATION

1. **Spinning Section** (lower) which provide gyroscopic stability and is mounted with solar panels.
2. **A despun section** (Upper) does not spin (or spins very slowly or intermittently) to point antenna and/or other sensors in a specific direction

## Despun section

Rotates, too, at a much slower rate of one rotation per orbit (day)—keeping the antennas pointed at the earth and preventing the satellite from going into a flat spin (which is the natural tendency).



## Spinning section

Provides basic stabilization and can rotate as fast as 100 RPM (in the case of the early GOES satellites)

# SPIN STABILIZED SATELLITE WITH SOLAR PANNELS

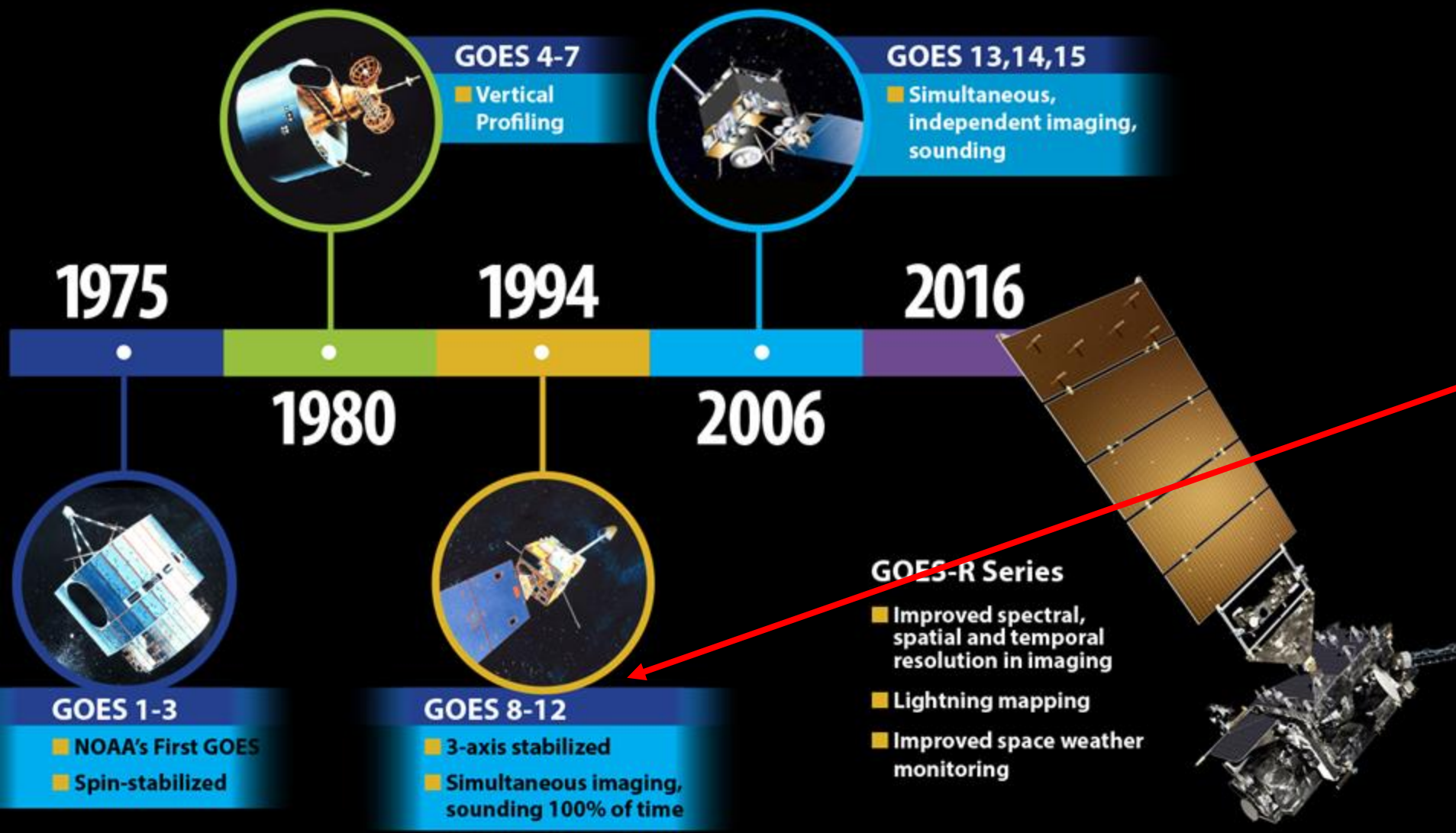


## Despun section

Rotates at a much slower rate of one rotation per orbit day,

## Spinning section

Provides basic stabilization .



**Dual Stabilization**  
 From 1994, GOES satellites designed by the National Oceanic and Atmospheric Administration (USA) stopped using dual stabilization and commenced using three-axis stabilization.

GOES stands for Geostationary Operational Environmental Satellite.

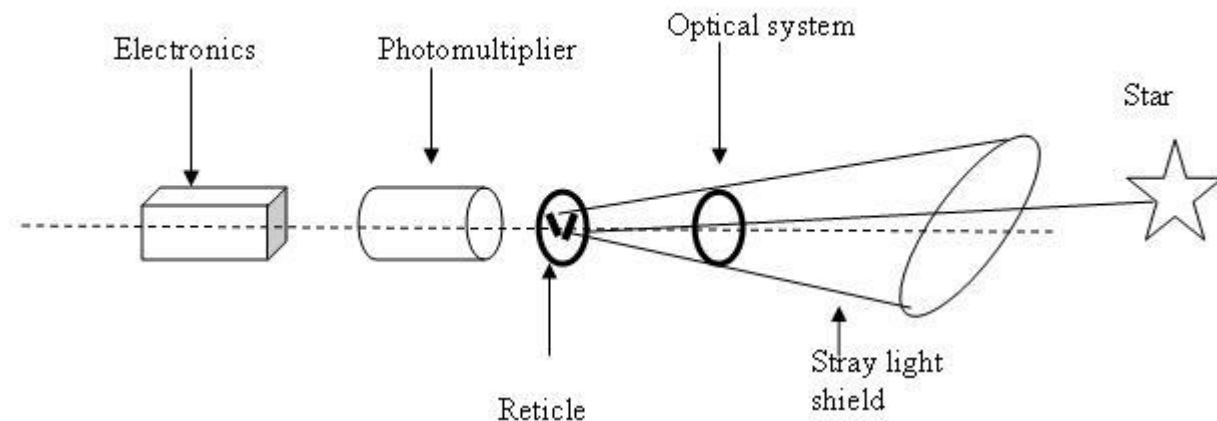
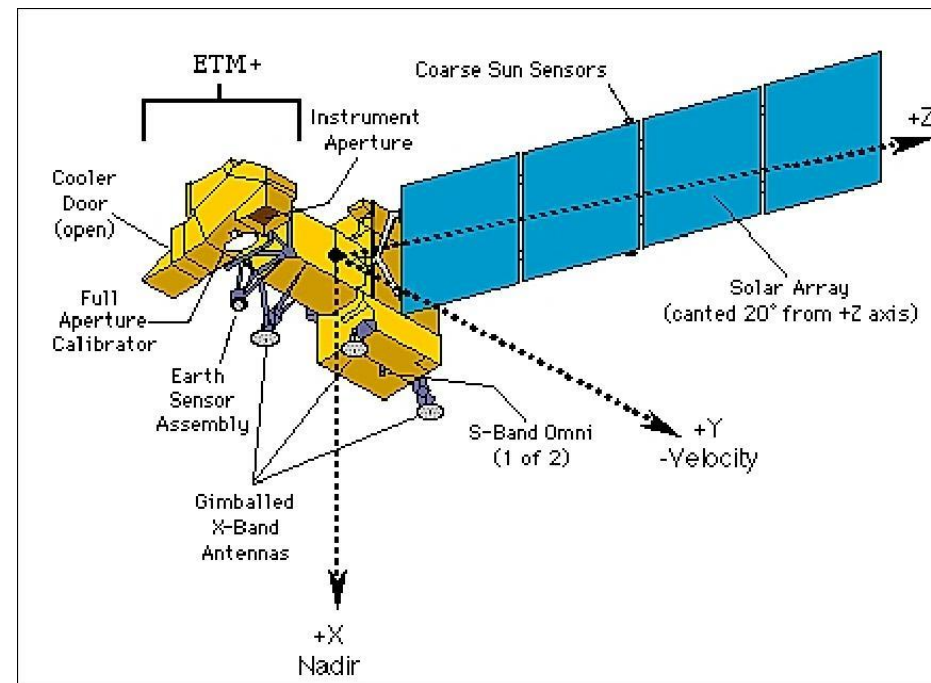
## 2. 3 ATTITUDE AND ORBIT SYSTEM

1. **Attitude** of a satellite refers to its orientation in space so that:
  - a) the antennas point their narrow beam towards desired areas on the earth, and
  - b) the solar panels point to the sun.
2. Forces that tend to alter the attitude of the satellite include:
  - (a) Asymmetry of the earth's gravitational field;
  - (b) Gravitational forces of the sun and moon;
  - (c) Pressure from sun radiation;
  - (d) Magnetic field of the earth.

## 2.4 ATTITUDE CONTROL - SENSORS

Three types of sensors are used in a satellite system, i.e

- 1. Earth Sensors:** Senses the direction to Earth. They are usually an infrared cameras. Used for attitude control, antenna pointing and gyro updating.
- 2. Sun Sensors:** Devices used to detect the position of the sun. Sun sensors are used for attitude control, solar array pointing, gyro updating, and fail-safe recovery.
- 3. Star Sensors:** star coordinates in the spacecraft frame and provide attitude information when these observed coordinates are compared with known star directions obtained from a star catalogue.



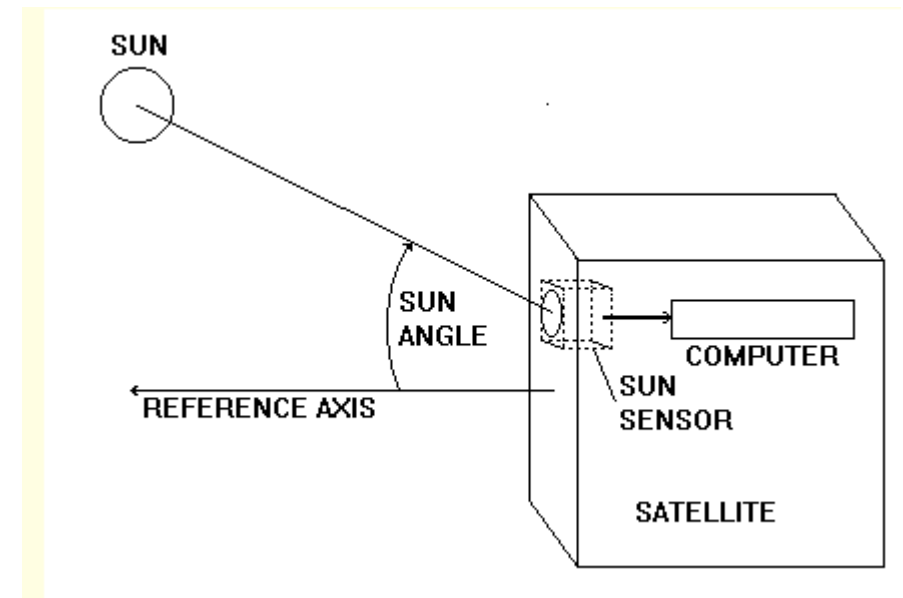
# 2.5 EARTH SENSORS

## Earth Sensors on Dual-Satellites

1. Consist of passive infra-red sensors in the 14-16  $\mu\text{m}$  wavelength which senses infrared rays coming from earth.
2. Two earth sensors positioned at  $5^\circ$  north and  $5^\circ$  south of the spin axis.
3. An on-board circuit measures the phase difference between the two sensors and sends the information (earth aspect angle) through the telemetry system to the earth control station

## 2.6 SUN SENSORS

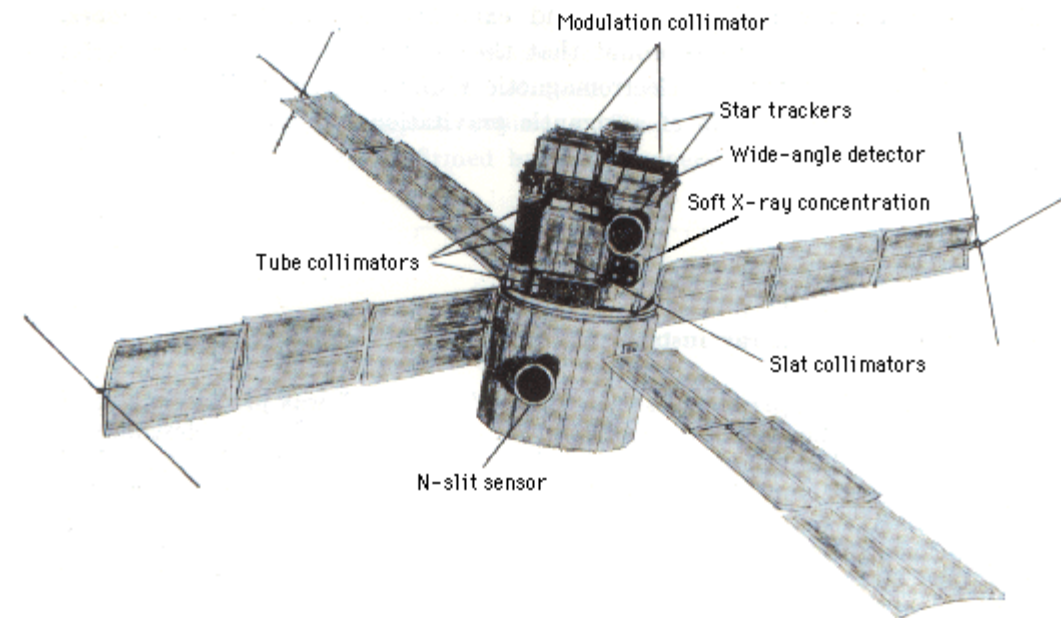
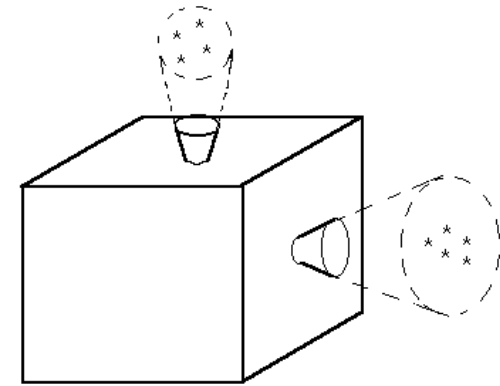
1. The sun-sensor measures **angle between "sun line" and a reference axis** in the satellite.
2. **Sun Sensor uses two sensors** operating in the visible spectrum as shown.
3. One sensor is **mounted parallel to the spin axis** while the other is **canted 35°**.
4. Data from the on-board processor is **transmitted to the earth control station to determine the solar aspect angle**.



## 2.7 STAR SENSORS

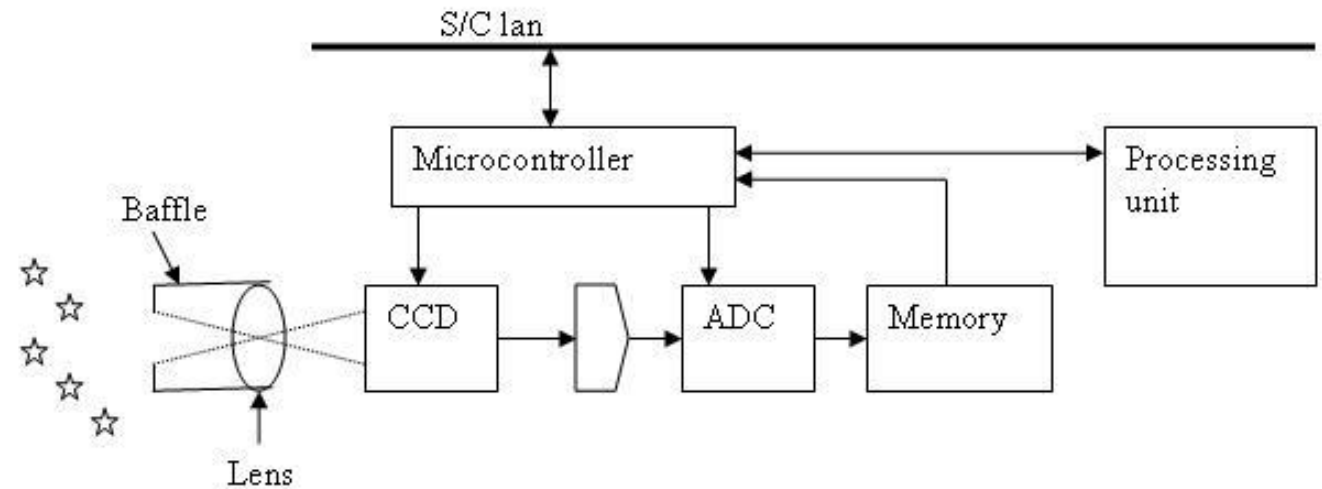
### Star sensor

1. Cameras mounted on the satellite in different directions used to identify different star patterns.
2. On-board computer compares camera images with stored star catalogue to determine which way the satellite is oriented.



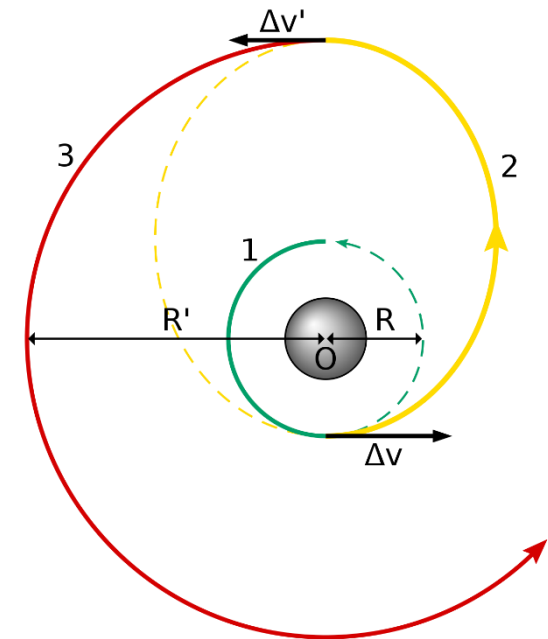
# 2.7.1 STAR SENSOR COMPONENTS

- 1. Star tracking system** consists of the following:
  - a) The star tracker control
  - b) Star image processing
  - c) Star pattern identification
  - d) Attitude estimation
- 2. CCD or “charge couple devices” star trackers** are the most popular nowadays because in comparison with the other sensors they have the highest value for the quantum efficiency parameter while they also cover a wide wavelength region.



# 3.0 PROPULSION SYSTEM

1. The propulsion system is an **on-board reaction control system used to generate forces on the satellite** when orbit and attitude control is required.
2. The propulsion system uses two types of actuators:
  - (a) **Low thrust actuators** – used for attitude and orbit corrections with velocity corrections of less than 50 m/s
  - (b) **High thrust actuators** – used for orbit transfers such as the geostationary orbit injection from the transfer orbit.



# 3.1 LOW-THRUST ACTUATORS

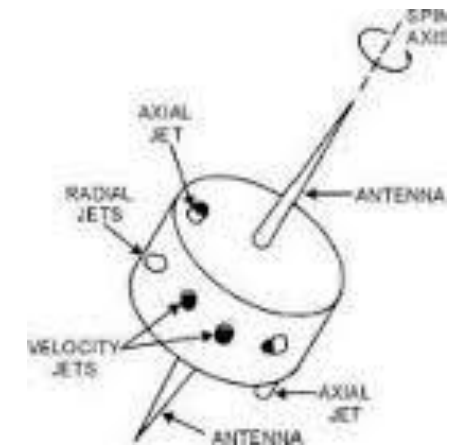
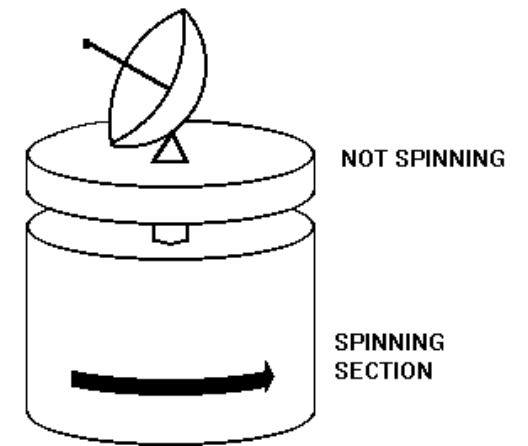
1. The low-thrust actuators can be further classified as:
  - (a) **Chemical thrusters** which generate gases through chemical reaction of propellants. The gases are then released through nozzles.
  - (b) **Electric thrusters** which provide thrust by accelerating ionized mass in an electromagnetic field.
2. **Communication satellites usually use Chemical thrusters.**

## 3.2 ATTITUDE CONTROL-DUAL SPIN REVISITED

1. In order to control attitude of a satellite in space, the satellite must be properly oriented and stabilized.
2. Two common methods of stabilization are:

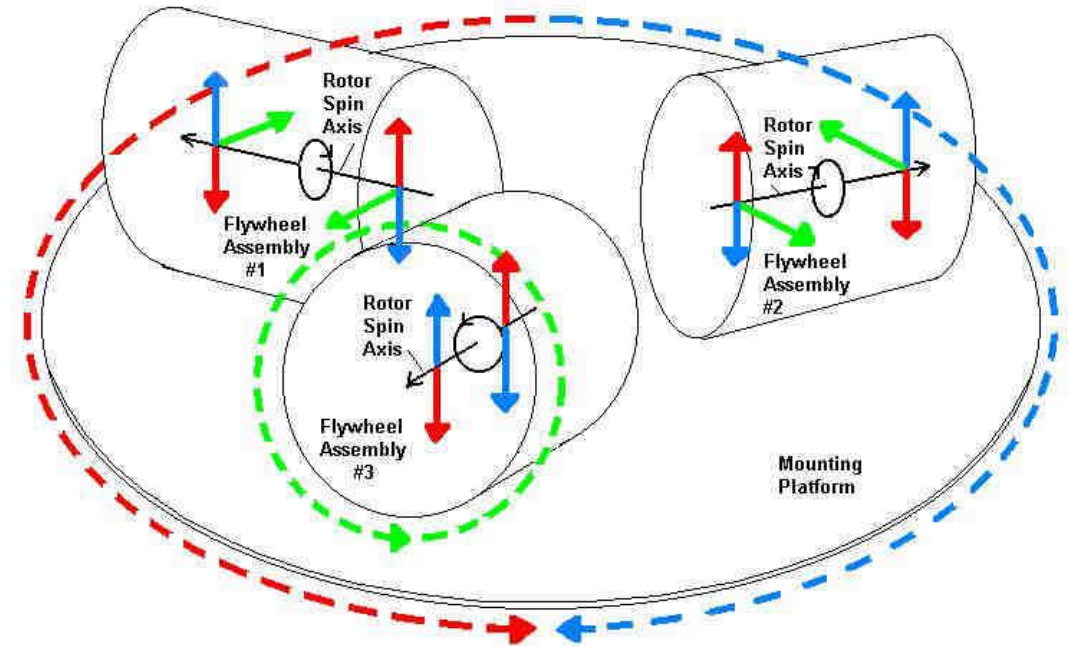
### (a) Spin stabilization

- i. The spacecraft is rotated at 30-60 rotations per minute.
- ii. The spin provides a powerful gyroscopic action that maintains the axis in a correct direction.
- iii. The antenna is usually mounted at one end of the drum and spun in the opposite direction to that of the drum (despun).
- iv. The despun section is kept stationary by counter rotation.



# 3.3 THREE AXIS BODY STABILIZATION

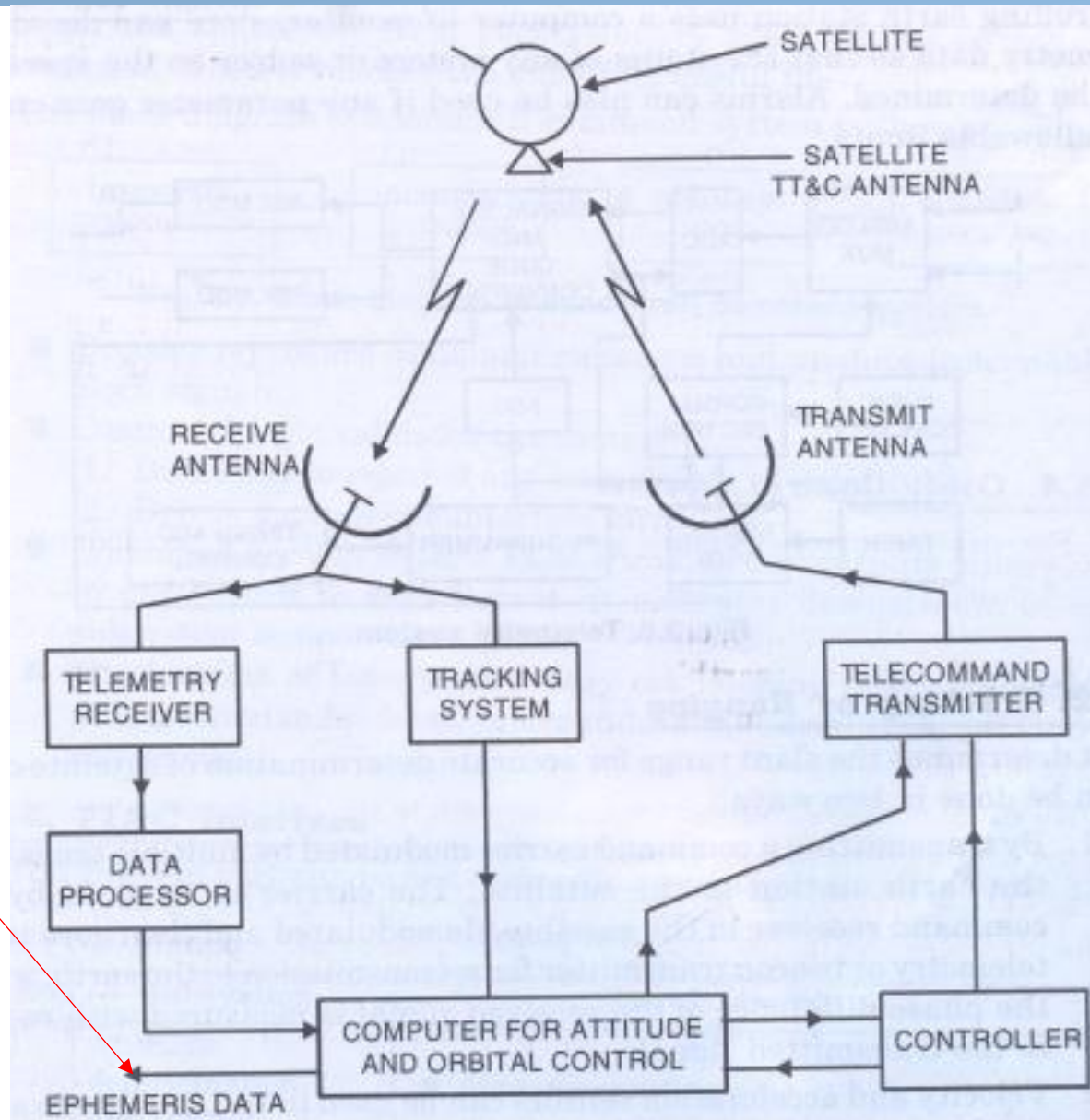
1. Three axis stabilization is achieved by rotating it round three axes, i.e yaw, roll and pitch.
2. Three momentum wheels are mounted on three perpendicular axis corresponding to yaw, roll and pitch.
3. A momentum wheel is a high-speed wheel driven by a motor and kept in a sealed chamber.
4. Changes in attitude are measured by the sensors and transmitted to the earth control station.
5. The data is analysed by the ground station and commands are sent to increase or decrease the speed of the momentum wheels.



# 4.0 TELEMETRY, TRACKING AND COMMAND

1. The Telemetry, Tracking and Control (TT&C) subsystem **provides vital control and monitoring function required to maintain the functionality of a satellite.**
2. TT&C is the **only way to observe and to control the satellite's functions and condition** from the ground.
3. TT& C functions include:
  - a) **Carrier tracking**
  - b) **Command reception and detection**
  - c) **Telemetry modulation and transmission**
  - d) **Ranging**
  - e) **Subsystem operation control**

# 4.1 BLOCK DIAGRAM OF TT&C SYSTEM



## **Ephemeris data**

Data file giving the calculated positions of the satellite at regular intervals throughout a period.

## 4.2 TELEMETRY SECTION

1. **The telemetry section** collects data from many sensors on the satellite and sends data to the controlling earth station.
2. The information transmitted includes:
  - a) The pressure in the fuel tank
  - b) Voltage and currents in the power conditioning unit
  - c) Current drawn by each subsystem
  - d) Critical voltages and currents in the communication subsystem
  - e) Temperature in the subsystems
  - f) Attitude
  - g) Position of various switches
  - h) Typically 100 sensors are required to monitor the above.

## 4.3 TRACKING AND RANGE

Slant range of a satellite can be accurately determined by using two methods:

- (a) By transmitting a command modulated carrier which is received by the satellite transponder and retransmitted back to the earth after a specific delay. Same methods as that used in radar DME.
- (b) Use of velocity and acceleration sensors mounted on the satellite to sense change in the orbital position. The earth based controller station receives these data through telemetry and after analysis relays orbit correction.

# GROUND-BASED SOLUTIONS

